

ABSTRACT**FIG. 4**

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BLADE ARRANGEMENT FOR GAS TURBINE ENGINE

A blade arrangement for a gas turbine engine includes a plurality of blades mounted for rotation on a disc so as 10 to extend radially therefrom and a retention member, the retention member including an attachment portion which is attached to the disc and an abutment portion for resisting forward axial movement of at least one of the blades relative to the disc. The blade arrangement further 15 comprises restraint means spaced from the attachment portion of the retention member, for substantially preventing radially outward movement of the abutment portion of the retention member when a forward axial force is applied by the blade to the abutment portion.

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